

EASA	NOTIFICATION OF A PROPOSAL TO ISSUE AN AIRWORTHINESS DIRECTIVE	
	PAD No.: 14-058 Date: 25 March 2014 Note: This Proposed Airworthiness Directive (PAD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.	
	In accordance with the EASA Continuing Airworthiness Procedures, the Executive Director is proposing the issuance of an EASA Airworthiness Directive (AD), applicable to the aeronautical product(s) identified below. All interested persons may send their comments, referencing the PAD Number above, to the e-mail address specified in the 'Remarks' section, prior to the consultation closing date indicated.	
Design Approval Holder's Name: AIRBUS		Type/Model designation(s): A330 and A340 aeroplanes
TCDS Numbers: EASA.A.004 and EASA.A.015		
Foreign AD: Not applicable		
Supersedure: None		
ATA 57		
Wings – Center Wing Box Fastener Holes at Frame 40 Vertical Web – Inspection		
Manufacturer(s): Airbus (formerly Airbus Industrie)		
Applicability: Airbus A330-201, A330-202, A330-203, A330-223, A330-223F, A330-243, A330-243F, A330-301, A330-302, A330-303, A330-321, A330-322, A330-323, A330-341, A330-342 and A330-343 aeroplanes, all manufacturer serial numbers, except those on which Airbus modification (Mod) 55792 or Mod 55306 has been embodied in production, and except those on which Airbus Repair Instruction R57115092 has been embodied in service on both Right Hand (RH) and Left Hand (LH) sides. Airbus A340-211, A340-212, A340-213, A340-311, A340-312 and A340-313, aeroplanes, all manufacturer serial numbers except those on which Airbus Mod 55792 or Mod 55306 has been embodied in production, and except those on which Airbus Repair Instruction R57115092 has been embodied in service on both RH and LH sides.		
Reason: During accomplishment of A330 Airworthiness Limitation Item (ALI) task 57-11-04 on the rear fitting of the Frame (FR) 40 between stringers 38 and 39 on both LH/RH sides, cracks were found on an adjacent hole. After reaming at second oversize of the subject hole, the crack was still present. Other crack findings on this adjacent hole have been reported on A330 and A340-200/-300 aeroplanes as a result of sampling inspections. This condition, if not detected and corrected, could affect the structural integrity of the aeroplane. For the reasons described above, this AD requires, removal of the fasteners		

	and repetitive rototest inspections of fastener holes at FR40 vertical web located above Center Wing Box (CWB) lower panel reference and/or below CWB lower panel reference on both sides and, depending on findings, accomplishment of the applicable corrective actions. Note: These holes affected by this AD are different from the ones affected by EASA AD 2009-0001.																											
Effective Date:	[TBD: 14 days after final AD issue date]																											
Required Action(s) and Compliance Time(s):	Required as indicated, unless accomplished previously: (1) Within the compliance time defined in Table 1 of this AD, as applicable, and thereafter, at intervals not to exceed the values defined in Airbus Service Bulletin (SB) A330-57-3114, or SB A330-57-3115, or SB A330-57-3116, or SB A340-57-4123, or SB A340-57-4124 Revision 01, or SB A340-57-4125, as applicable, depending on aeroplane configuration and utilisation, remove the fasteners and accomplish a special detailed inspection of fastener holes at FR40 vertical web, on both sides, located above and below CWB lower panel reference in accordance with the instructions of applicable Airbus SB, as specified in Table 2 of this AD, depending on aeroplane configuration. Table 1 – Initial Inspection <table><tr><th colspan="2">Compliance time (whichever occurs later, A or B)</th></tr><tr><td>A</td><td>Within the threshold defined in Airbus SB A330-57-3114, or SB A330-57-3115, or SB A330-57-3116, or SB A340-57-4123, or SB A340-57-4124 Revision 01, or SB A340-57-4125, as applicable, depending on aeroplane configuration and utilisation.</td></tr><tr><td>B</td><td> <u>For A330 aeroplanes:</u> Within 2 400 Flight Cycles (FC) or 24 months, whichever occurs first after the effective date of this AD. <u>For A340 aeroplanes:</u> Within 1 300 FC or 24 months, whichever occurs first after the effective date of this AD. </td></tr></table> Table 2 – Applicable Airbus SB <table><tr><th>Aeroplane Configuration</th><th>Airbus SB</th><th>Location to be Inspected</th></tr><tr><td>A330-300 pre-mod 44360</td><td>A330-57-3114</td><td>Below</td></tr><tr><td>A340-200/-300 pre-mod 44360</td><td>A340-57-4123</td><td>Below</td></tr><tr><td>A330-200/-300 post-mod 44360 and pre-mod 49202</td><td>A330-57-3116</td><td>Below</td></tr><tr><td>A340-200/-300 post-mod 44360 and pre-mod 49202</td><td>A340-57-4125 Revision 01</td><td>Below</td></tr><tr><td>A330-200/-300 pre-mod 55306 and A330-200/-300 pre-mod 55792</td><td>A330-57-3115</td><td>Above</td></tr><tr><td>A340-200/-300 pre-mod 55306 and A340-200/-300 pre-mod 55792</td><td>A340-57-4124</td><td>Above</td></tr></table>	Compliance time (whichever occurs later, A or B)		A	Within the threshold defined in Airbus SB A330-57-3114, or SB A330-57-3115, or SB A330-57-3116, or SB A340-57-4123, or SB A340-57-4124 Revision 01, or SB A340-57-4125, as applicable, depending on aeroplane configuration and utilisation.	B	<u>For A330 aeroplanes:</u> Within 2 400 Flight Cycles (FC) or 24 months, whichever occurs first after the effective date of this AD. <u>For A340 aeroplanes:</u> Within 1 300 FC or 24 months, whichever occurs first after the effective date of this AD.	Aeroplane Configuration	Airbus SB	Location to be Inspected	A330-300 pre-mod 44360	A330-57-3114	Below	A340-200/-300 pre-mod 44360	A340-57-4123	Below	A330-200/-300 post-mod 44360 and pre-mod 49202	A330-57-3116	Below	A340-200/-300 post-mod 44360 and pre-mod 49202	A340-57-4125 Revision 01	Below	A330-200/-300 pre-mod 55306 and A330-200/-300 pre-mod 55792	A330-57-3115	Above	A340-200/-300 pre-mod 55306 and A340-200/-300 pre-mod 55792	A340-57-4124	Above
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B	<u>For A330 aeroplanes:</u> Within 2 400 Flight Cycles (FC) or 24 months, whichever occurs first after the effective date of this AD. <u>For A340 aeroplanes:</u> Within 1 300 FC or 24 months, whichever occurs first after the effective date of this AD.																											
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A340-200/-300 pre-mod 55306 and A340-200/-300 pre-mod 55792	A340-57-4124	Above																										

	(2) If, during any inspection as required by paragraph (1) of this AD, no crack is detected, before next flight, install new fasteners in transit fit in accordance with the instructions of Airbus SB A330-57-3114 or SB A330-57-3115 or SB A330-57-3116 or SB A340-57-4123 or SB A340-57-4124 Revision 01 or SB A340-57-4125, as applicable to aeroplane type and model. (3) If, during any inspection as required by paragraph (1) of this AD, a crack is detected, before next flight, accomplish the applicable corrective actions (additional inspection followed by repetitive inspections or contact Airbus to get further instructions) in accordance with the instructions of Airbus SB A330-57-3114 or SB A330-57-3115 or SB A330-57-3116 or SB A340-57-4123 or SB A340-57-4124 Revision 01 or SB A340-57-4125, as applicable to aeroplane type and model. (4) Installation of new fasteners as required by paragraph (2) of this AD does not constitute terminating action for the repetitive inspections required by paragraph (1) of this AD. (5) Accomplishment of the corrective actions as required by paragraph (3) of this AD does not constitute terminating action for the repetitive inspections required by paragraph (1) of this AD. (6) Accomplishment of the initial and repetitive inspections as required by this AD cancels the need to accomplish Airworthiness Limitation Items tasks 57-11-04 and 57-11-02. (7) Inspections and corrective actions, accomplished before the effective date of this AD in accordance with the instructions of Airbus Technical Disposition Reference LR57D11023270 or LR57D11029171 or LR57D11029173 or LR57D11030741 or LR57D11029170, LR57D11023714, LR57D11029172 or LR57D11030740 or SB A340-57-4124 at original issue, are acceptable to comply with the initial inspections and corrective actions as required, respectively, by paragraphs (1) and (3) of this AD. After the effective date of this AD, inspections and corrective actions must be accomplished as required by this AD.
Ref. Publications:	Airbus SB A330-57-3114 original issue dated 12 March 2013. Airbus SB A330-57-3115 original issue dated 04 April 2013. Airbus SB A330-57-3116 original issue dated 12 March 2013. Airbus SB A340-57-4123 original issue dated 12 March 2013. Airbus SB A340-57-4124 original issue dated 04 April 2013 or Revision 01 dated 22 August 2013. Airbus SB A340-57-4125 original issue dated 12 March 2013. The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.
Remarks:	1. This Proposed AD will be closed for consultation on 22 April 2014. 2. Enquiries regarding this PAD should be referred to the Safety Information Section, Executive Directorate, EASA. E-mail: ADs@easa.europa.eu. 3. For any question concerning the technical content of the requirements in this PAD, please contact: Airbus – Airworthiness Office – EIAL; E-mail: airworthiness.A330-A340@airbus.com.