EASA	AIRWORTHINESS DIRECTIVE
	AD No.: 2014-0272 **Date: 12 December 2014** Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

This AD is issued in accordance with EU 748/2012, Part 21.A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].

Design Approval Holder's Name:	**Type/Model designation(s):**
AIRBUS	A300, A300-600 and A300-600ST aeroplanes

TCDS Numbers:	EASA.A.172 and EASA.A.014
Foreign AD:	Not applicable
Supersedure:	None

ATA 57	**Wings – Frame 40 Lower Junction Fastener Holes – Inspection / Repair**
Manufacturer(s):	Airbus (formerly Airbus Industrie)
Applicability:	Airbus A300 and A300-600 aeroplanes, all models, all Manufacturer Serial Numbers (MSN), and Airbus A300-600ST aeroplanes, all MSN.
Reason:	Following the A300-600 Extended Service Goal (ESG2) exercise, specific inspections for cracks were performed in fittings of frame (FR) 40, in areas not covered by any existing task. Findings were identified on an A300-600 aeroplane withdrawn from service in the lower tension bolt area at rib one junction (both sides). This condition, if not detected and corrected, could lead to crack initiation, affecting the structural integrity of the aeroplane. To address this potential unsafe condition, an inspection programme was developed for the fitting around the fastener holes located at FR40 lower wing junction, left hand (LH) and right hand (RH) sides. For the reasons described above, this AD requires repetitive High Frequency Eddy Current (HFEC) inspections and rototest inspections of the fitting around the fastener holes located at FR40 lower wing junction and, depending on findings, accomplishment of a repair.
Effective Date:	29 December 2014

TE.CAP.00110-004 © European Aviation Safety Agency. All rights reserved.
Proprietary document. Copies are not controlled. Confirm revision status through the EASA-Internet/Intranet.

Required Action(s) and Compliance Time(s):	Required as indicated, unless accomplished previously: (1) Within 1 000 flight hours (FH) after the effective date of this AD, and, thereafter, at intervals not to exceed 1 000 FH, accomplish a HFEC inspection of fasteners 1 to 3 at FR 40 lower junction, LH and RH sides, and of the fitting around the fastener holes in accordance with the instructions of Airbus Service Bulletin (SB) A300-57-0257, SB A300-57-6115 or SB A300-57-9030, as applicable to aeroplane model. (2) If, during any HFEC inspection as required by paragraph (1) of this AD, any crack is found, before next flight, contact Airbus for approved repair instructions and accomplish those instructions accordingly. (3) Within 36 months after the effective date of this AD, remove fasteners 1 to 3 at FR40 lower junction, LH and RH sides, measure the diameter of the fastener holes and, before next flight, accomplish the actions specified in Table 1 of this AD, as applicable depending on measurement results, in accordance with the instructions of Airbus SB A300-57-0257, SB A300-57-6115 or SB A300-57-9030: Table 1: Actions following Measurement of Fastener Holes (4) Accomplishment of a rototest inspection on an aeroplane, as required by paragraph (3) of this AD, constitutes terminating action for the repetitive HFEC inspections as required by paragraph (1) of this AD for that aeroplane. (5) If, during the rototest inspection as required by paragraph (3) of this AD, any crack is found, before next flight, contact Airbus for approved repair instructions and accomplish those instructions accordingly. (6) If, during the rototest inspection as required by paragraph (3) of this AD, no crack is found, before next flight, install new fasteners of the same diameter in special clearance fit for fasteners 1 to 3 at FR40 lower junction, LH and RH sides, and, thereafter, at intervals not to exceed 7 000 flight cycles, repeat the rototest inspection in accordance with the instructions of Airbus SB A300-57-0257, SB A300-57-6115 or SB A300-57-9030. (7) If, during any rototest inspection as required by paragraph (6) of this AD, any crack is found, before next flight, contact Airbus for approved repair instructions and accomplish those instructions accordingly.
Ref. Publications:	Airbus SB A300-57-0257 original issue, dated 04 April 2014. Airbus SB A300-57-6115 original issue, dated 04 April 2014. Airbus SB A300-57-9030 original issue, dated 31 March 2014. The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.

Measurement Results	Corrective Actions
One or more hole diameters are outside the tolerance of the nominal diameter and are outside the tolerance of the first and second oversize.	Contact Airbus for approved repair instructions and accomplish those instructions accordingly.
All hole diameters are within the tolerance of the nominal diameter or the first or second oversize.	Accomplish a rototest inspection of the fastener holes at FR40 lower junction, LH and RH sides.

TE.CAP.00110-004 © European Aviation Safety Agency. All rights reserved.
Proprietary document. Copies are not controlled. Confirm revision status through the EASA-Internet/Intranet.

Remarks:	1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD. 2. This AD was posted on 30 April 2014 as PAD 14-077 for consultation until 28 May 2014. The Comment Response Document can be found at http://ad.easa.europa.eu. 3. Enquiries regarding this AD should be referred to the Safety Information Section, Certification Directorate, EASA. E-mail: ADs@easa.europa.eu. 4. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS SAS – EIAW (Airworthiness Office) E-mail: continued.airworthiness-wb.external@airbus.com.

TE.CAP.00110-004 © European Aviation Safety Agency. All rights reserved.
Proprietary document. Copies are not controlled. Confirm revision status through the EASA-Internet/Intranet.