

EASA	NOTIFICATION OF A PROPOSAL TO ISSUE AN AIRWORTHINESS DIRECTIVE	
	PAD No.: 14-132 Date: 20 August 2014 Note: This Proposed Airworthiness Directive (PAD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.	
	In accordance with the EASA Continuing Airworthiness Procedures, the Executive Director is proposing the issuance of an EASA Airworthiness Directive (AD), applicable to the aeronautical product(s) identified below. All interested persons may send their comments, referencing the PAD Number above, to the e-mail address specified in the 'Remarks' section, prior to the consultation closing date indicated.	
Design Approval Holder's Name: AIRBUS HELICOPTERS		Type/Model designation(s): SA 341 G, SA 342 J helicopters
TCDS Number: EASA.R.125		
Foreign AD: Not applicable		
Supersedure: None		
ATA 65	Main Rotor – Main Rotor Hub – Torsion Bars - Inspection	
Manufacturer(s):	Aerospatiale	
Applicability:	SA 341 G and SA 342 J helicopters, all manufacturer serial numbers.	
Reason:	Several cases of cracks were found on the polyurethane (PU) coating of the Main Rotor Hub (MRH) torsion bars P/N 704A33633274 installed on military SA 341 helicopters. These parts can also be installed on civilian SA 341 and SA 342 helicopters. In addition, analysis of the cracked torsion bars has shown that small areas of superficial corrosion on the strands inside the bars can also develop during the manufacturing process. The cracking of the PU coating near these areas and the associated penetration of water can lead to further and deeper development of the corrosion. This condition, if not detected and corrected, allows water to penetrate into the torsion bar, causing corrosion and failure of the metal strands inside the bar, which could lead to torsion bar failure, resulting in in-flight loss of a main rotor blade and consequent loss of control of the helicopter. To address this unsafe condition, Airbus Helicopters (AH) issued Alert Service Bulletin (ASB) n° SA341/SA342-05.40, providing instructions for reducing the intervals of the existing repetitive inspection of the torsion bars, which are described in the applicable maintenance manual (document identified as MDE) and currently required through the existing instructions for continued airworthiness (document identified as PRE). For the reasons described above, this AD requires reduced intervals for the	

	repetitive inspection of the torsion bars compared to the existing intervals indicated in the current PRE document.																		
Effective Date:	[TBD: 14 days after final AD issue date]																		
Required Action(s) and Compliance Time(s):	Required as indicated, unless accomplished previously: (1) Initially, within the compliance time specified in Table 1 or Table 2 of this AD, as applicable, and, thereafter, at intervals not to exceed the values as specified in Table 3 of this AD, as applicable, inspect the MRH torsion bars P/N 704A33633274 in accordance with the instructions of AH MDE Work Card n° 65.12.607. Table 1 – Initial Inspection of Torsion Bars installed <u>less</u> than 5 years <table> <tr> <th>Flight Hours (FH) accumulated by the helicopter (on the effective date of this AD) since last inspection of the torsion bar</th><th>Compliance Time</th></tr> <tr> <td><u>less</u> than 320 FH</td><td>Before exceeding 420 FH or 24 months, whichever occurs first since last inspection</td></tr> <tr> <td>320 FH or more</td><td>Within 100 FH, or within the scheduled inspection time limit for the torsion bars as indicated in the current PRE, whichever occurs first after the effective date of this AD.</td></tr> </table> Table 2 – Initial Inspection of Torsion Bars installed 5 years or more <table> <tr> <th>Service life (FH or calendar time) accumulated by the helicopter on the effective date of this AD since last inspection of the torsion bar</th><th>Compliance Time</th></tr> <tr> <td>less than 320 FH, <u>and</u> last inspection performed <u>less</u> than 6 months before the effective date of this AD</td><td>Before exceeding 420 FH or 12 months, whichever occurs first since last inspection</td></tr> <tr> <td>320 FH or more, <u>or</u> last inspection performed 6 months or more before the effective date of this AD</td><td>Within 100 FH, or within 6 months, or within the scheduled inspection time limit for the torsion bars as indicated in the current PRE, whichever occurs first after the effective date of this AD.</td></tr> </table> Table 3 – Repetitive Inspections of Torsion Bars <table> <tr> <th>Duration of installation of the torsion bar on a helicopter</th><th>Inspection Intervals (not to exceed)</th></tr> <tr> <td><u>less</u> than 6 years</td><td>420 FH or 24 months, whichever occurs first, until reaching the limit of 6 years after first installation on a helicopter</td></tr> <tr> <td>6 years or more</td><td>420 FH or 12 months, whichever occurs first, without exceeding the existing service life limits indicated in the airworthiness limit section of the current PRE.</td></tr> </table>	Flight Hours (FH) accumulated by the helicopter (on the effective date of this AD) since last inspection of the torsion bar	Compliance Time	<u>less</u> than 320 FH	Before exceeding 420 FH or 24 months, whichever occurs first since last inspection	320 FH or more	Within 100 FH, or within the scheduled inspection time limit for the torsion bars as indicated in the current PRE, whichever occurs first after the effective date of this AD.	Service life (FH or calendar time) accumulated by the helicopter on the effective date of this AD since last inspection of the torsion bar	Compliance Time	less than 320 FH, <u>and</u> last inspection performed <u>less</u> than 6 months before the effective date of this AD	Before exceeding 420 FH or 12 months, whichever occurs first since last inspection	320 FH or more, <u>or</u> last inspection performed 6 months or more before the effective date of this AD	Within 100 FH, or within 6 months, or within the scheduled inspection time limit for the torsion bars as indicated in the current PRE, whichever occurs first after the effective date of this AD.	Duration of installation of the torsion bar on a helicopter	Inspection Intervals (not to exceed)	<u>less</u> than 6 years	420 FH or 24 months, whichever occurs first, until reaching the limit of 6 years after first installation on a helicopter	6 years or more	420 FH or 12 months, whichever occurs first, without exceeding the existing service life limits indicated in the airworthiness limit section of the current PRE.
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	(2) If, during any inspection as required by paragraph (1) of this AD, a crack in the PU coating of a torsion bar is found which matches or exceeds the criteria defined in MDE Work Card n° 65.12.607, before next flight, replace the affected torsion bar with a serviceable one in accordance with the instructions of AH ASB n° SA341/SA342-05.40. (3) Replacement of a torsion bar on a helicopter as required by paragraph (2) of this AD does not constitute terminating action for the repetitive inspections as required by paragraph (1) of this AD for that helicopter. (4) From the effective date of this AD, installation of a MRH torsion bars P/N 704A33633274 is allowed, provided that, following installation, MRH torsion bars P/N 704A33633274 is inspected and corrected as required by this AD.
Ref. Publications:	Airbus Helicopters ASB n° SA341/SA342-05.40 original issue dated 28 April 2014. Airbus Helicopters MDE Work Card n° 65.12.607. The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.
Remarks:	1. This Proposed AD will be closed for consultation on 17 September 2014. 2. Enquiries regarding this PAD should be referred to the Safety Information Section, Executive Directorate, EASA. E-mail: ADs@easa.europa.eu. 3. For any question concerning the technical content of the requirements in this PAD, please contact: Airbus Helicopters (STDI) – Aéroport de Marseille Provence 13725 Marignane Cedex, France, Telephone +33 (4) 42 85 97 97; Fax +33 (4) 42 85 99 66, E-mail: Directive.technical-support@eurocopter.com.