EASA	NOTIFICATION OF A PROPOSAL TO ISSUE AN AIRWORTHINESS DIRECTIVE
	PAD No.: 14-134 **Date: 02 September 2014** Note: This Proposed Airworthiness Directive (PAD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

In accordance with the EASA Continuing Airworthiness Procedures, the Executive Director is proposing the issuance of an EASA Airworthiness Directive (AD), applicable to the aeronautical product(s) identified below.
All interested persons may send their comments, referencing the PAD Number above, to the e-mail address specified in the 'Remarks' section, prior to the consultation closing date indicated.

Design Approval Holder's Name:	**Type/Model designation(s):**
AIRBUS	A380 aeroplanes

TCDS Number:	EASA.A.110
Foreign AD:	Not applicable
Supersedure:	None

ATA 53	**Fuselage – Section 19 Upper Cross Fitting Upper Beam Inner Cap Joints – Inspection**
Manufacturer(s):	Airbus
Applicability:	Airbus A380-841, A380-842 and A380-861 aeroplanes, all manufacturer serial numbers (MSN).
Reason:	A refined analysis in correlation with the results of a fatigue test accomplished on the A380 Frame (FR) 108 Upper Beam identified a structural area for which the inspection threshold and intervals published in the applicable Airworthiness Limitations Section (ALS) Part 2 must be reduced to retain the structural integrity of the affected area. Failure to accomplish the affected inspections within the reduced threshold and intervals could result in an unsafe condition. To address this potentially unsafe condition, Airbus issued Service Bulletin (SB) A380-53-8074 to provide the relevant inspection instructions. For the reasons described above, this AD requires repetitive special detailed inspections of the Section 19 upper cross fitting upper beam inner cap joints, both left hand (LH) and right hand (RH) sides, and, depending on findings, accomplishment of applicable corrective actions. Accomplishment of these inspections also cancels the need for the affected A380 ALS Part 2 tasks.
Effective Date:	[TBD: 14 days after final AD issue date]

Required Action(s) and Compliance Time(s):	Required as indicated, unless accomplished previously: (1) Within the compliance time as defined in Table 1 of Appendix 1 of this AD, as applicable, and, thereafter, at intervals not to exceed the values as defined in Table 2 or Table 3 of Appendix 1 of this AD, as applicable, accomplish one of the actions as specified in paragraph (1.1) or (1.2) of this AD on the LH and RH sides of the Section 19 upper cross fitting upper beam inner cap joint in accordance with the instructions of Airbus SB A380-53-8074: (1.1) Accomplish a High Frequency Eddy-Current (HFEC) inspection of the upper beam and extension beam inner cap area at cross fitting joint positioned forward of FR108, or (1.2) Accomplish a rototest of the sixth fastener hole of the lower attachment row at the upper cross fitting and the upper forward beam inner cap forward of FR108 area (refer to Figure A-SSAAA of Airbus SB A380-53-8074), and a HFEC of the rest of locations. (2) If, during any rototest as specified in paragraph (1.2) of this AD, the first oversized fastener installation has to be exceeded at location of the sixth fastener hole of the lower attachment row at the upper cross fitting and the upper forward beam inner cap positioned forward of FR108 area, before next flight, contact Airbus for further instructions and accomplish those instructions accordingly. (3) If, during any HFEC inspection or rototest as required by paragraph (1) of this AD, any damage is detected, before next flight, contact Airbus for approved repair instructions and, within the compliance time defined in those instructions, accomplish the repair accordingly. (4) Inspections accomplished before the effective date of this AD in accordance with the instructions of Technical Disposition (TD) TD_G1_S4_04931_2013 issue C or TD_G1_S4_05440_2013 Issue A or TD_G1_S4_05583_2013 issue A are acceptable to comply with the initial action as required by paragraph (1) of this AD. After the effective date of this AD, all actions must be accomplished in accordance with the instructions of Airbus SB A380-53-8074. (5) Compliance with the requirements of this AD cancels the need to accomplish A380 ALS, Part 2 – Airworthiness Limitation Item, task 535100-00012-01A and task 535100-00026-01A.
Ref. Publications:	Airbus SB A380-53-8074 at original issue dated 30 July 2014. The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD. Airbus TD_G1_S4_04931_2013 Issue C dated 23 October 2013. Airbus TD_G1_S4_05440_2013 Issue A dated 06 November 2013. Airbus TD_G1_S4_05583_2013 issue A dated 21 November 2013.
Remarks:	1. This Proposed AD will be closed for consultation on 16 September 2014. 2. Enquiries regarding this PAD should be referred to the Safety Information Section, Certification Directorate, EASA. E-mail: ADs@easa.europa.eu. 3. For any question concerning the technical content of the requirements in this PAD, please contact: AIRBUS SAS - EIANA (Airworthiness Office), E-mail: account.airworth-A380@airbus.com.

Appendix 1

Table 1 – Inspection Threshold

Aeroplane MSN	Compliance Time
0005, 0007, 0009, 0011, 0013, 0015, 0017, 0020, 0022, 0023, 0025, 0026, 0027, 0028,0029, 0030, 0033, 0046 and 0051	Within 28 000 FH or 3 800 FC, whichever occurs first after the aeroplane first flight, but not later than 24 months after the effective date of this AD
0006, 0008, 0010, 0012, 0016, 0019 and 0021	Within 31 200 FH or 3 500 FC, whichever occurs first after the aeroplane first flight, but not later than 24 months after the effective date of this AD
All other MSN	Within 23 500 FH or 3 200 FC, whichever occurs first after the aeroplane first flight

Table 2 – Repetitive inspection for aeroplanes which have been previously inspected in accordance with paragraph (1.1) of this AD

Aeroplane configuration	Compliance Time (FH or FC, whichever occurs first since last inspection)
Aeroplanes not modified in accordance with Airbus modification (mod) 64090 or aeroplanes on which fastener ASNA2392-8-07 has been installed at the opportunity of the previous inspections accomplished as specified by paragraph (1.2) of this AD	Within 9 200 FH or 1 200 FC
Aeroplanes modified in accordance with Airbus mod 64090 or aeroplanes on which fastener EN6115K8-7 or EN6115K8X7 has been installed at the opportunity of the previous inspections accomplished as specified by paragraph (1.2) of this AD	Within 8 000 FH or 1 050 FC

Table 3 – Repetitive inspections for aeroplanes which have been previously inspected as required by paragraph (1.2) of this AD

Compliance Time
Within 13 300 FH or 1 800 FC, whichever occurs first since last inspection